1.13) In a proposed jet propulsion system for an automobile, air is drawn in vertically through a large intake in the roof at a rate of 3 kg/s, the velocity through this intake being small. Ambient pressure and temperature are 100 kPa and  $30^{\circ}$  C, respectively. This air is compressed and heated and then discharged horizontally out of a nozzle at the rear of the automobile at a velocity of 500 m/s and a pressure of 140 kPa. If the rate of heat addition to the air stream is 600 kW, find the nozzle discharge area and the thrust developed by the system. **Solution:** 

Given: air is drawn vertically,  $\dot{m} = 3$  kg/s,  $V_1 \approx 0$  m/s,  $p_{amb} = p_0 = 100$  kPa,  $T_{amb} = T_0 = 30^{\circ}$  C = 303 K

air is compressed and heated and discharged horizontally,

 $V_e = 500 \,\mathrm{m/s}, \, p_e = 140 \,\mathrm{kPa}, \, \dot{q} = 600 \,\mathrm{kW}.$ 

To calculate:  $A_e = ?$ , Thrust =?.

The schematic diagram of the problem description is shown in Fig. 1.

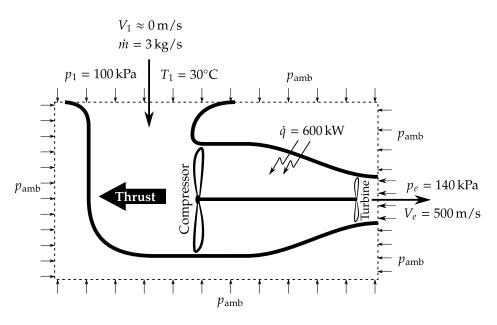


Fig. 1: Schematic diagram for problem description

Applying the conservation of energy through the system,

Rate of energy entering + Rate of heat addition = Rate of energy leaving + Work done by the system

$$\dot{m} c_p T_0 + \dot{q} = \dot{m} c_p T_e + \dot{m} \frac{V_e^2}{2} + W$$
$$3 \times 1005 \times 303 + 600 \times 10^3 = 3 \times 1005 \times T_e + 3 \times \frac{500^2}{2} + 0$$
$$T_e = \frac{3 \times 1005 \times 303 + 600 \times 10^3 - 3 \times \frac{500^2}{2}}{3 \times 1005} = 377.62686567 \,\mathrm{K}$$

The density at the exit can be calculated using the ideal gas equation,

$$\rho_e = \frac{p_e}{R T_e} = \frac{140 \times 10^3}{287 \times 377.6} = 1.29176 \, \text{kg/m}^3$$

Applying the conservation of mass equation,

$$A_e = \frac{\dot{m}}{\rho_e V_e} = \frac{3}{1.29176 \times 500}$$
$$A_e = 0.0046448 \,\mathrm{m}^2 \,\mathrm{.}$$

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Applying the conservation of horizontal momentum,

Thrust = 
$$\dot{m}_e V_e + (p_e - p_{amb}) A_e$$
  
Thrust =  $3 \times 500 + (140 \times 10^3 - 100 \times 10^3) \times 0.0046448$   
Thrust =  $1685.792 \text{ N}$ .